Advanced Mirror Technology Development (AMTD) for Very Large Space Telescopes

COR Program Technology Development

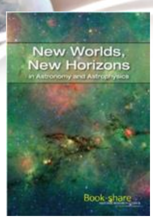
Annual Progress Presentation

PI: H. Philip Stahl, MSFC

Optical Fabrication & Testing 2017

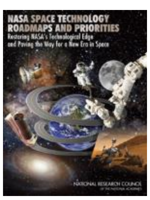
1

Space Telescopes require Mirror Technology



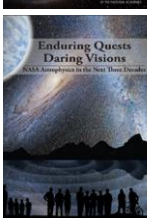
Astro2010 Decadal Study recommended technology development (page 7-17) for a potential future:

- Exoplanet Mission (New-Worlds Explorer)
- UVOIR Space Telescope (4 meter or larger)



2012 NASA Space Technology Roadmaps & Priorities: Top Technical Challenge C2 recommended:

- New Astronomical Telescopes that enable discovery of habitable planets, facilitate advances in solar physics, and enable the study of faint structures around bright objects …



2014 Enduring Quests Daring Visions recommended:

- LUVOIR Surveyor with sensitivity to locate the bulk of planets in the solar neighborhood and reveal the details of their atmospheres.

2

Objective

AMTD's objective is to mature to towards TRL-6 the critical technologies needed to produce 4-m or larger flight-qualified UVOIR mirrors by 2018 so that a viable mission can be considered by the 2020 Decadal Review.

AMTD is not developing technology for a specific mission, but two 'Decadal' missions that benefit from AMTD are:

- **Habitable Exoplanet Imager (HabEx) and**
- **Large UVOIR Surveyor (LUVOIR).**

3

Multiple Technology Paths

Just as JWST's architecture was driven by launch vehicle, future mission's architectures (mono, segment or interferometric) will depend on capacities of future launch vehicles (and budget).

Since we cannot predict future, we must prepare for all futures.

To provide the science community with options, we are pursuing multiple technology paths for both monolithic and segmented aperture telescopes.

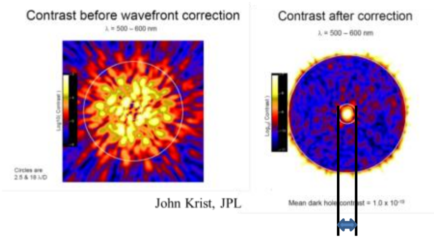
All potential UVOIR mission architectures (monolithic, segmented or interferometric) share similar mirror needs:

- Very Smooth Surfaces — < 10 nm rms
- Thermal Stability — Low CTE Material
- Mechanical Stability — High Stiffness Mirror Substrates

4

'The' System Challenge: Dark Hole

- Imaging an exoplanet, requires blocking 10^{10} of host star's light
- An internal coronagraph (with deformable mirrors) can create a 'dark hole' with < 10^{-10} contrast.



John Krist, JPL

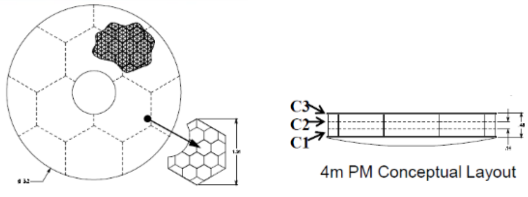
- **Ultra-smooth, Ultra-Stable Mirror Systems are critical to achieving and maintaining the 'dark hole'**

Krist, Trauger, Unwin and Traub, "End-to-end coronagraphic modeling including a low-order wavefront sensor", SPIE Vol. 8422, 844253, 2012; doi: 10.1117/12.927143
Shaklan, Green and Palacios, "TPFC Optical Surface Requirements", SPIE 626511-12, 2006.

5

Large Stable Mirror Substrates

- Future large-aperture space telescopes (regardless of monolithic or segmented) need ultra-stable mechanical and thermal performance for high-contrast imaging.
- This requires larger, thicker, and stiffer substrates.
- Phase 1 demonstrated stacked core low-temperature fusion process to cost effectively make mirrors thicker than 300 mm by making a 40 cm 'cut-out' of a 4-m mirror.
- Phase 2 demonstrated lateral scaling of the stacked core process by making a 1.5 m subscale of a 4-m mirror.

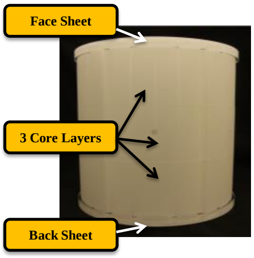


4m PM Conceptual Layout

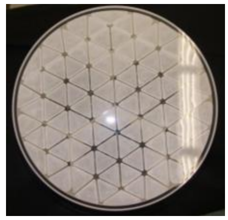
43 cm Deep Core Mirror

Harris successfully demonstrated 5-layer ‘stack & fuse’ technique which fuses 3 core structural element layers to front & back faceplates.

43 cm ‘cut-out’ of a 4 m dia, > 0.4 m deep, 60 kg/m^2 mirror substrate.



Post-Fusion Side View
3 Core Layers and Vent Hole Visible



Post-Fusion Top View
Pocket Milled Faceplate

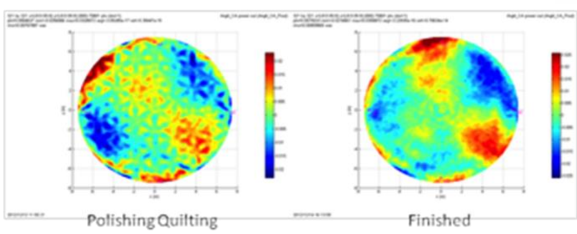
Post Slump:
2.5 meter Radius of Curvature

This technology advance leads to stiffer 2 to 4 to 8 meter class substrates at lower cost and risk for monolithic or segmented mirrors.

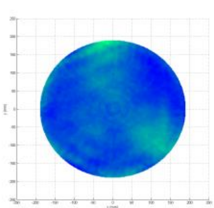
Matthews, Gary, et al, *Development of stacked core technology for the fabrication of deep lightweight UV quality space mirrors*, SPIE Conference on Optical Manufacturing and Testing X, 2013.

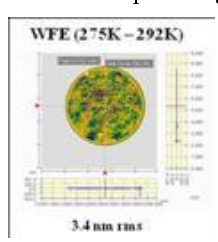
Mid/High Spatial Frequency Error

- Harris polished 43 cm deep-core mirror to a zero-gravity figure of 5.5 nm rms using ion-beam figuring to eliminate quilting.



- MSFC tested 43 cm mirror from 250 to 300K. Its thermal deformation was insignificant (smaller than 4 nm rms ability to measure the shape change)

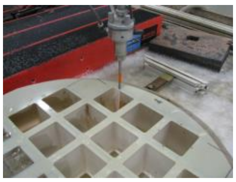




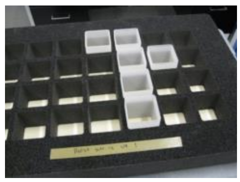
8

Strength Testing

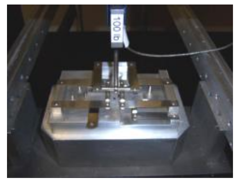
- AMTD-1: Harris strength tested the core to core LTF bond strength on 12 Modulus of Rupture (MOR) test articles.
  - Weibull 99% survival value was 15% above conservative design allowable. Data ranged from 30% to 200% above design allowable.
- AMTD-2: A-Basis test of core rib to core rib LTF bond strength.
  - 60+ MOR Samples: 30+ samples aligned; 30+ core misaligned
  - A-basis Weibull 99% confidence strength allowable for 49 samples is 17.5MPa; ~50% higher than the strength of core-to-plate LTF bonds.



MOR Boxes in Abrasive Water Jet (AWJ)



post AWJ, pre-LTF assembly



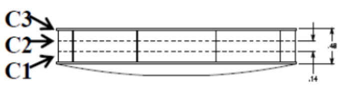
MOR sample in Test Fixture

9

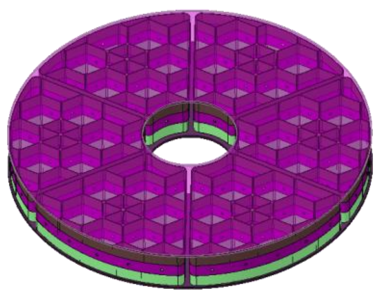
Phase 2: Demonstrates Lateral Scaling

Designed 4-m x 500 mm on-axis mirror then scale down to 1.5 m x 185 mm.

- (2) ULE® face plates
- (3) ULE® glass boules

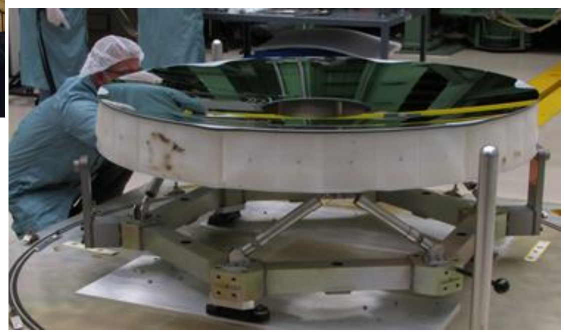


4m PM Conceptual Layout



10

1.5-m x 185 mm 450 Hz ULE® Mirror



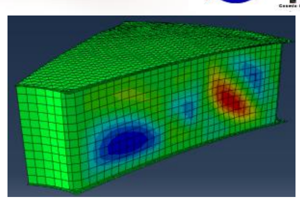
Photo's courtesy of Harris Corporation

11

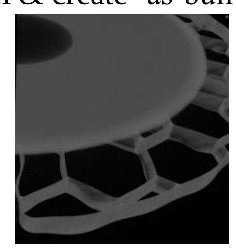
Visco-Elastic Behavior

Non-linear visco-elastic modeling predicted Wall Bowing.

Mirror was designed to accommodate predicted bowing.



X-Ray Computed Tomography used to quantify internal structure to correlated with visco-elastic model & create 'as-built' STOP model.



Lessons Learn have been documented.

12

1.5-m ULE® Mirror Status

Next is Thermal Performance characterization testing.

Given the importance of mid-spatial frequency errors (both static and dynamic) in producing the ‘dark-hole’, AMTD will quantify:

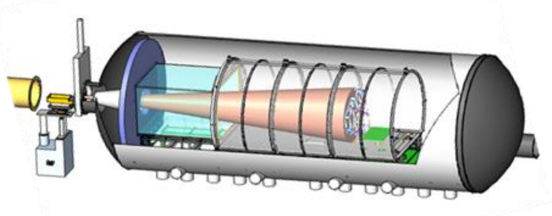
- Thermal induced quilting.
- CTE variation induced Surface Figure Error
- Surface Thermal Stability

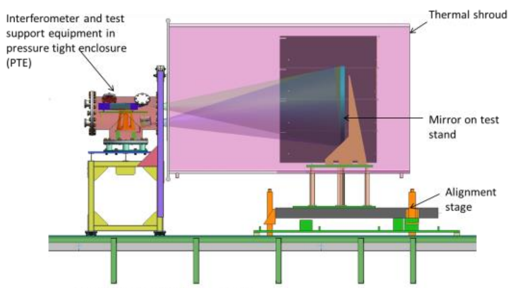
AMTD did this for the Schott 1.2m Extreme-Lightweight Zerodur Mirror (ELZM).

AMTD also predicted and quantified ELZM static and dynamic mechanical performance (gravity sag and first mode frequency).

13

MSFC Thermal-Optical Test Capabity





AMTD-2 test configuration with PTE

Test Measured Data at 250K

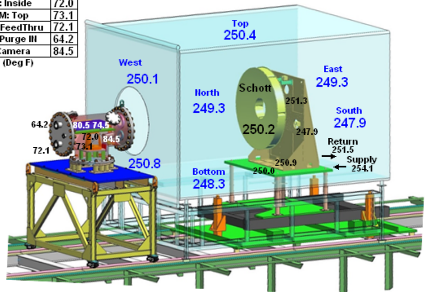
09/16/16 08:10:57

AMTD2 / Schott Cryo Test

PTE

PhaseCam East	74.5
PhaseCam West	80.5
PTE: Inside	72.0
ADM: Top	73.1
Cable FeedThru	72.1
PTE: Purge IN	64.2
IR Camera	84.5

(Deg F)



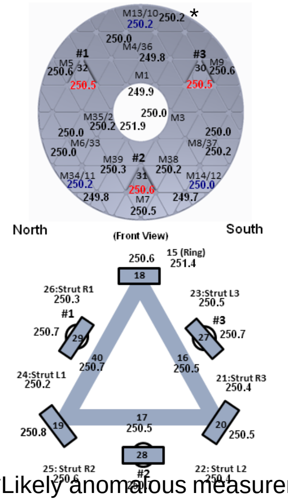
Shroud

Top	250.4
North	249.3
South	247.9
Bottom	248.3
West Top	250.1
West Bottom	250.8
East	249.3

(Kelvin)

	Shroud		Schott	
Average	249.4	K	250.2	K
Rate	-0.1	K/HR	-0.1	K/HR
Max	250.8	K	251.9	K
Min	247.9	K	249.7	K
Grad	3.0	K	2.2	K

ΔT~0.8K



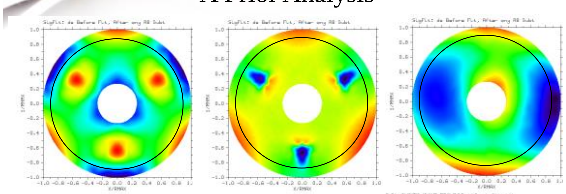
M1- Top Hole	249.9
M2 - North Hole	251.9
M3 - South Hole	250.0
M4 - 12:00	250.0
M5 - 10:00	250.6
M6 - 8:00	250.0
M7 - 6:00	250.5
M8 - 4:00	250.2
M9 - 2:00	250.3
M10- Top Edge	250.2
M11 - 8:00 Edge	249.8
M12 - 4:00 Edge	249.7
M13 - Top Front	250.2
M14 - 4:00 Front	250.0
M33 - 8:00 (w/M6)	250.0
M34 - 8:00 (w/M11)	250.2
M35 - 8:00 (w/M2)	250.2
M36 - 12:00 (w/M4)	249.8
M37 - 4:00 (w/M8)	250.0
M38 - 5:00	250.2
M39 - 7:00	250.3
30 - South Pad	250.5
31 - Bottom Pad	250.6
32 - North Pad	250.5
15 - 12:00 Ring	251.4
16 - Delta_3	250.5
17 - Delta_2	250.5
18 - Top Bracket	250.6
19 - South Bracket	250.8
20 - North Bracket	250.5
21 - Strut R3	250.4
22 - Strut L2	250.4
23 - Strut L3	250.5
24 - Strut L1	250.2
25 - Strut R2	250.6
26 - Strut R1	250.3
27 - South Mount	250.7
28 - Bottom Mount	250.7
29 - North Mount	250.7
[illegible] - Delta_1	[illegible]

(Kelvin)

*Likely anomalous measurement ignored

Correlated Model with Measured for 294K to 250K

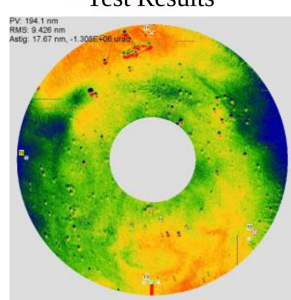
A Prior Analysis



Thermal Gradients (1.28 nm RMS) | Mount Effects (0.81 nm RMS) | Inhomogeneity* (9.55 nm RMS)

* Random CTE map was generated with Schott specified 5 ppb/K PV homogeneity.

Test Results



Measured SFE (9.4 nm RMS)

- **CTE drives thermal performance.**
- **Model accuracy depends on CTE knowledge.**

Quilting

While the cryo-deformation phase maps show negligible quilting associated with the mechanical structure of the mirror substrate, there is ‘fringe print through’.

The ‘fringe print-through’ is caused by two factors:

- Mirror surface figure is ~400 nm PV
  - Gravity Sag ~ 300 nm Astigmatism PV
  - Zero-G Figure ~ 115 nm PV
- The PhaseCAM uses a 4-bucket algorithm.

A known feature of the 4-bucket algorithm is that if the phase-shift is not exact, there is a ‘ghost’ pattern in the phase map with spatial frequency 2X that of the fringes.

Conclusions

- AMTD uses science-driven systems engineering to derive performance specifications from science requirements then define & execute a long-term strategy to mature technologies to enable future large aperture space telescopes.
- Because we cannot predict the future, we are pursuing multiple technology paths including monolithic & segmented mirrors.
- AMTD Phase 2:
  - Fabricate ⅓-scale model of a 4-m x 400-mm class ~150 Hz ULE® mirror (1.5-m x 185-mm 450 Hz).
  - Characterize optical performance of two candidate lightweight primary mirrors from 250K to ambient: 1.2-m ELZM and 1.5-m ULE.
  - Correlate Integrated Modeling Tools
- Lessons Learned from the 1.5m ULE mirror have been documented.